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MECHANICAL PROPERTIES, INCLUDING FRACTURE TOUGHNESS AND FATIGUE, CORROSION CHARACTERISTICS AND FATIGUE-CRACK PROPAGATION RATES OF STRESS-RELIEVED ALUMINUM ALLOY HAND FORGINGS

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Contract No. F33615-68-C-1385 Project No. 7381

Third Technical Management Report August 15, 1968 - November 15, 1968

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ABSTRACT

Except for a few tests, the tensile, compressive, shear and bearing properties have been determined for all but three of a total of forty 2014-T652, 2024-T852, 7075-T7352 and 7079-T652 hand forgings scheduled for test. Ratios among these properties have been calculated. Plane-strain fracture-toughness values have been determined for two of the hand forgings. The results of axial-stress fatigue tests (R=0.0) of smooth specimens are presented.

The preparation of specimens for the stress-corrosion and exfoliation tests has been initiated, and exposure of the specimens will be started soon.

Tests of the 6x24-in. 2014-T652 hand forging are in progress to study the effects of notch geometry, specimen length and stress on the rate of fatigue crack propagation.

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Third Technical Management Report

MECHANICAL PROPERTIES, INCLUDING FRACTURE TOUGHNESS AND FATIGUE CORROSION CHARACTERISTICS AND FATIGUE-CRACK-PROPAGATION RATES OF STRESS-RELIEVED ALUMINUM ALLOY HAND FORGINGS

I. Introduction.

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The design mechanical properties, fracture toughness, corrosion characteristics and fatigue-crack propagation rates are four of the most important factors involved in the selection and efficient design of aircraft structures. Such data are needed for aluminum alloy hand forgings for several reasons:

(1) much of the published design data has become obsolete by a change in the basis of specifying minimum properties, from one in which the length, width and thickness were considered, to one where only the thickness is involved; (2) the development of a technique of stress relieval by cold work in compression has resulted in relatively new tempers (TX52) for many of the alloys; and (3) there have been some significant problems with forged parts in recent years that were related to fracture and stress-corrosion characteristics.

Accordingly, the properties of hand forgings of several aluminum alloys currently being used in aircraft structures are being determined under this contract. The tests are intended to provide statistically reliable data for deriving design mechanical properties for MIL-HDBK-5A, including stress-strain and compressive tangent-modulus curves. In addition, data concerning the fracture toughness, axial-stress fatigue, stress-corrosion, exfoliation and fatigue-crack propagation rates are being obtained.

This Third Technical Management Report summarizes the results of tests carried out during the third quarter of the contract, and the general status of the program at this time.

II. Material.

All but one of the hand forging samples to be investigated have now been obtained. The 6x24-in. 7075-T/352 forging which had been received did not meet the tentative minimum tensile properties and was returned to the plant for reheat treatment. The reheat-treated sample is expected within the next month.

III. Procedure.

All the specimens and test procedures were described in the First Technical Management Report, dated May 15, 1968.

IV. Progress During Quarter.

A. <u>Mechanical Properties</u>

A.l. Tensile, Compressive, Shear and Bearing

Tensile, compressive, shear and bearing tests have been made of 37 forgings, the results of which are shown in Tables I through IV. The ratios showing the relationships among these mechanical properties are shown in Table V.

Since the data for the various sizes of hand forgings of any one alloy are not complete, no detailed analysis has been made of the test results.

A.2. Fracture Toughness

Notch-bend fracture tests have been made of two of the hand forgings. The average values obtained are as follows:

Alloy		Sample		
and Temper	Cross-Sect. Size, in.	Number	Direction*	K_{Ic} , psi \sqrt{in} .
2014-1652	5 x 20	341013	LW WL	29 200 19 600
7075-17352	4 x 16	341030	LW WL	32 700 26 500

The above values are considered to be valid, although for some individual specimens the stress intensity used in fatigue cracking was slightly in excess of 50 per cent of the $K_{\rm IC}$ value, and the fatigue crack deviated from straightness by slightly more than 5 per cent of the thickness.

Specimens from most of the other hand forging samples have been prepared and are in the process of being fatigue cracked.

A.3. Axial Stress Fatigue

The axial-stress fatigue (R=0.0) tests of specimens from all except the 6x12-in. 7075-T7352 hand forging have been started, and approximately 80 per cent of the scheduled tests have been completed. The results of the tests are plotted in Figs. 1 through 4.

^{*} The first letter indicates the direction of a line normal to the crack plane in the specimens; the second letter indicates the direction of crack growth. L - Longitudinal (major axis of forging); W - Width.

B. Corrosion Characteristics

Specimens have been obtained from sixteen of the twenty-three hand forgings scheduled for corrosion testing.

Specimens from the 2x8-in., 3x12-in. and 5x20-in. forgings have been machined, and it is expected that the stress-corrosion specimens will be exposed in late November, 1968. Specimens have also been prepared for exposure to the accelerated exfoliation test; however, due to the extensive backlog of specimens for this exposure, there will be some delay before the tests are started. Some of the specimens from the 4 and 6-in. thick samples have been prepared, but the testing is being delayed until all of the specimens from a given size of forging can be exposed concurrently.

Machined slices of the 6x6-in. 7075-T7352 and 7079-T652 forgings have been obtained for macroetching to determine the grain orientation. Preparation of specimens from these samples will be initiated shortly.

C. Fatigue Crack Propagation

Twenty long-transverse crack propagation specimens were prepared from the 6x24-in. 2014-T652 hand forging. Tests of these specimens have been initiated to study the effects of notch geometry, specimen length and periodic changes in stress on the rate of fatigue-crack propagation. The 6x24-in. 7079-T652 hand forging has been submitted to the Machine Shop for preparation of six additional specimens.

The program calls for evaluation of the effects of specimen orientation and humidity on the rate of crack propagation for two of the four alloys. It was originally indicated that the selection of the two alloys might be made when data comparing the propagation behavior of all four alloys was available. However, with the delay in receiving the forgings, this is not feasible. Thus, it is proposed that these tests be made on the alloys expected to show the greatest difference in resistance to stress-corrosion cracking; i.e., 7075-T7352 and 7079-T652.

V. Summary.

The tensile, compressive, shear and bearing properties determined for 37 of the 40 hand forgings scheduled for test are shown in Tables I through IV. The tensile properties of the hand forgings meet the applicable specified minimum-property requirements shown in Table VI. Ratios among the properties are as shown in Table V.

Notch-bend fracture toughness tests were made of two hand forgings, the average $K_{\mbox{\footnotesize{IC}}}$ values are shown in Section IV, A.2 of the text.

Approximately 80 per cent of the axial-stress fatigue tests have been completed. The results of the tests are plotted in Figs. 1 through 4.

Specimens from the 2, 3 and 5-in, thick forgings have been prepared and the stress-corrosion and accelerated exfoliation tests are to be started soon. Although some of the

specimens from the 4 and 6-in. thick samples have already been prepared, testing is being delayed so that all of the specimens from a given size of forging can be exposed concurrently.

Machined slices from the 6x6-in. 7075-T7352 and 7079-T652 forgings have been obtained to determine the grain orientation.

Tests of the 6x24-in. 2014-T652 hand forging have been initiated to study the effects of notch geometry, specimen length and stress on the rate of fatigue-crack propagation. It is planned to evaluate the effects of specimen orientation and humidity for alloys 7075-T7352 and 7079-T652; these two alloys would be expected to show the greatest difference in resistance to stress-corrosion cracking.

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VI. Tables and Figures.

I BALE I

WECHANICAL PROPERTIES OF STRESS-WELIEVED 2014-TARR ALUMINUM ALLOY HAND FORGINGS

(F33615-68-C-1385)

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TARLE I

MECHANICAL PROPERTIES OF STRESS-PFLIEVED 2024-TA52 ALUMINUM ALLOY MAND FORGINGS (F33615-6A-C-1384)

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TABLE 11

wechaniral properties of Staess-PFLIEVED 7075-17352 ALUMINUM ALLOY HAND FORGINGS

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		5.1	100 -7	41 600	6.3	0	69 300		1	-	!	!
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* OFFSET FOUALS 0.2 DFP CFNT † OFFSET FOUALS 2.0 DFD CFNT OF PIN DIAMFTER ‡ SPECIMENS AND FIXTUPFS CLEAMED ULTRASONICALLY IN TOSON § L* LONGITURINAL; LT* LONG TDANSVERSE; ST* SHORI TPANSVERSE

WFCHANICAL PROPERTIES OF STRESS-RELIEVED 7079-1652 ALUMINUM ALLOY HAND FORGINGS

(F33615-6A-C-1385)

* OFFSET FOUALS 0.2 PER FENT † OFFSET FOUALS 2.0 PER CENT OF PIN DIAMETER ‡ SPECIMENS AND FIXTHRES CLFANED ULTRASONICALLY IN TOSON § 1. LONGITHNINAL; LT. LONG TRANSVENSE; ST. SMORT TRANSVENSE

PATIOS AMONG THE TENSILE. COMPDESSIVE. SHFAP AND BFAPTUG PONDEPTIES
OF STRESS-BELIEVED ALUMINUM ALLOY MAND FORGINGS

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AND	SIZF TN.	NUMBER	TYSILI	TYS(LT)	TYS(ST)	15 (LT)	15([1)	15(1,7)	e/0=	e/D= e/D=	6 /D= 1.5	e /0=	6/0= 1.5	150	175(LT) 6/0= 6/0= 1.5 2.0	-0°2
2014-1652	a ×	341007	1.04	1.08	21.1	29.0	19.0	i	1.4.1	2.2	1.35	7.54	1.61	- x	1.34	1.54
	CIXC	341004	1.03	1.07	1.12	65.0	0.59	A.5.0	1.44	1.97	٦. ا	1.45	1.17	1.74	1.34	1.56
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2024-1852	۷ ×	341017	1.09	1.14	1.17	0.59	0.54	1	75.1	70.1	1.50	٠, د	1.31	1.74	1.40	1.79
	3x12	341018	1.05	90.0	1.12	0.5A	0.57	٥.55	1.29	1.67	1.34	۵۳.۱	* C	1.71	٠٠.1	7.67
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	64 A 6x12 6x24	341024 341025 341026	1.02	0.00	7.7.6	0.50	0.59	0.0	1.39	1.90	97.	1.40	1.3	1.79	1.43	3 6.
7075-17352	æ ~	341027	1.05	1.05	1.12	9.62	0.59	i	1.49	1.94	1.44	1.70	1.48	1.95	1.42	1.63
	3×1>	341028	1.01	1.10	1.14	92.0	0.40	0.60	1.44	<u>.</u>	\$.	1.74	- 7-	04.1	1.5	44.
	4 x 4 A	341029	1.05	1.09	41.1	0.60	0.10	0.00 a.v.o	1.44	2.00 1.84	42.1	1.73	1.5	2. 4 2. 4	3.6	1.47
	5x 5 5x10 5x20	341031 341032 341033	1.01	1.03 1.05	71.1	6.60	000	0 C C 4 4 A A	1.55	1.94 1.96	1.53	700	44.1	551	2.5	T I I
	6x 6 6x12 6x24	341034 341035 341036	1.04	1.92	=	99.0	0.63		٦٠,	7°°¢	P. 2.	ž .	1.53	10.7	1.57	1.84
7079~1652	2x A	341037	1.03	1.13	1.17	44.0	14.0	i	1.51	£0.5	1.53	1.74	1.50	3.	2.	1.75
	3412	341034	1.04	1.08	1.17	19.0	0.61	0.00	1.49	1.05	77.1	1.72	1.5.1	1.0	· ·	1.74
	A X A	341039 341040	1.03	1.10	1.16	0.63	0.62	0.50	1.44	2.9	1.50	١٠٢٠ ا	1.51	23.	10.1	1.77 1.48
	5x 5 5x10 5x20	341041 341042 341043	1.03	1.05	1.15	0.66 0.52 0.63	0.63	466	45.1 63.1	7.05 1.90	1 4 4 C 4 C	1.73	244.	551	1143	1.7.1
	6x12 6x12	341044 341045 341046		1.07	1.19	0.64	9.69	6.51	1.50	2.04	4.1	1.72	1.43	19.5	94.	1.76
								I Len								

SPECIPIED MINIMUM VALUES POR ALLMINUM ALLOY HAND PORGINGS (P37615-08-C-1765)

	Specification	\$ 44-x-5678	Moce	• ecc.	49-4-3576
	in ore	10044	•	#\@@@	I የጎዞጎምነምነ
Short-Transverse	Yleid Strength,* pri	8888 8888 8888	i	8888 8888 8888 8888 8888 8888 8888 8888 8888	10.000 10.000 10.000
chs	Strength,	6288 8888 8888	•	7888 7888	\$558 \$558 \$558
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Con Transiters	Strength.	20000 00000 00000	1	600	48238 88888 88888
	Tenalle Strength, S	<i>\$</i> 3338	!	\$	255@g
	15 40,	തനരപ്പ	. 1	rrm	ውውውውው
	Strength,	88.54.4.1.1. 0000000000000000000000000000000		4244 8888	\$\$\$\$\$\$ \$\$\$\$\$\$
	Tensile Strength,	**************************************		8888 8888	5555¢ 538838 58888
	Thi skness,	Banar	A11	Up thru 3.000 4.001-4.000 5.001-5.000 5.001-5.000	25.001-4.000 2.001-4.000 4.001-4.000 5.001-5.000 5.001-5.000
	A.1.2.3.	7emper 2014-1652	9024-T058	7075-11352	7079-168

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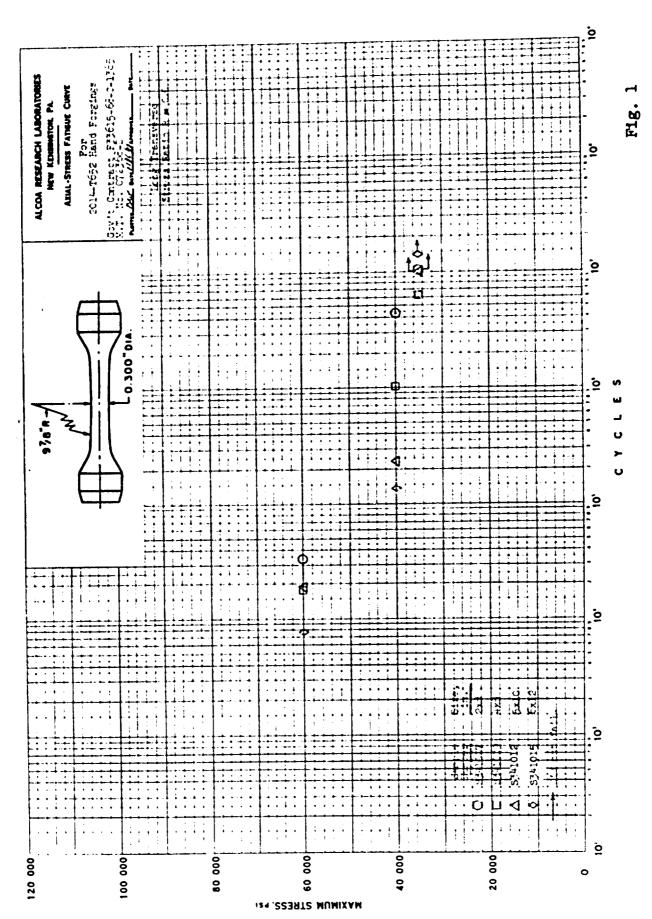
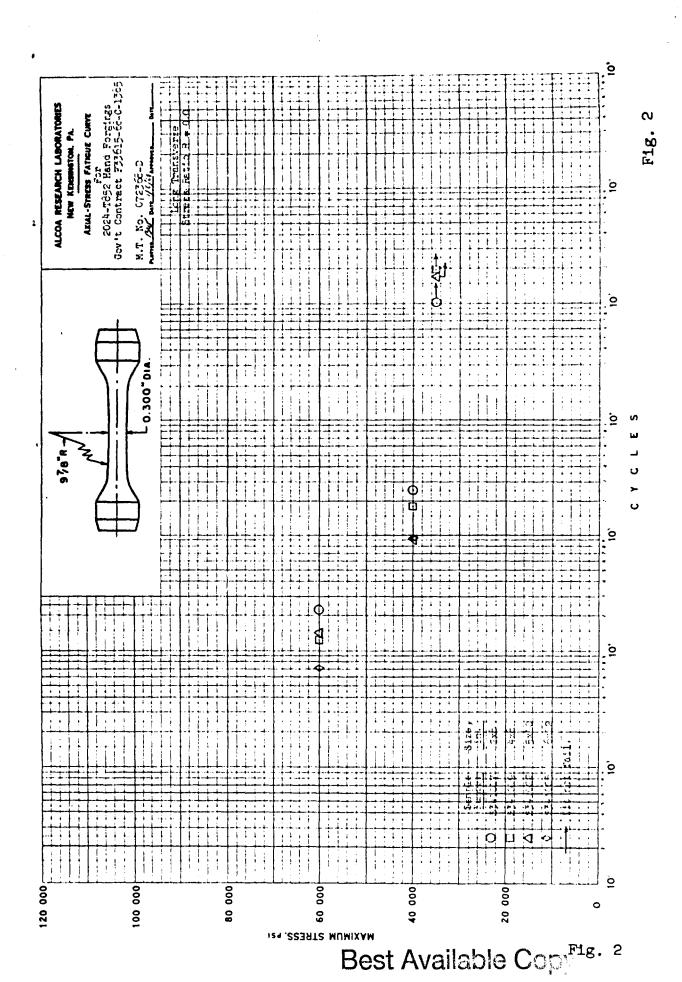
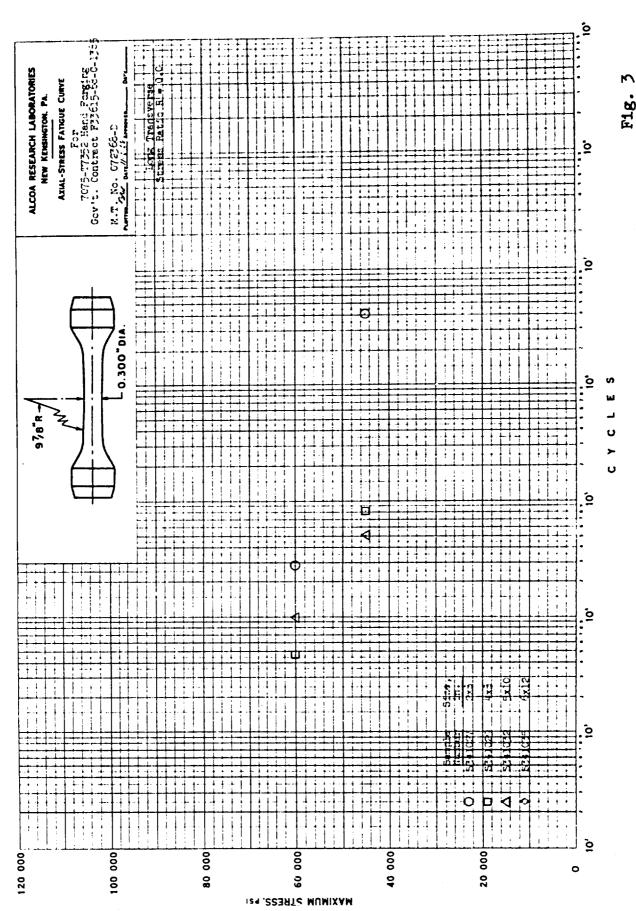


Fig. 1





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